

Sektion 2. Utlandstillverkad flygmateriel

TITEL: Kontroll av låsbricka i höjdstyrssystemet

GÄLLER: 369D, E, F och FF med installerad P/N 369D21800 eller P/N 369D21820 i höjdstyrningen.

ÅTGÄRD: Utför åtgärder angivna i MDHS SIN nr DN-185, EN-78 och FN-64 daterade 23 september 1994 eller senare utgåvor.

TID FÖR ÅTGÄRD: Inom 25 flygtimmar eller 90 dagar vilket som först inträffar räknat från detta LVD's utgivningsdatum.

UNDERLAG: FAA AD 94-24-04 (kopia bifogad)
MDHS Service Information Notice (SIN) nr DN-185, EN-78 och FN-64 daterade 23 september 1994 eller senare utgåvor.

REFERENS: FAA AD 94-24-04

UTGIVNINGS-DATUM: 1994-12-22

LFS: 1994:46

Åtgärd enligt LVD utgör nödvändig förutsättning för ifrågavarande flygmateriels luftvärdighet. Referens BCL M 1.11. Anteckning om åtgärd, som vidtagits i enlighet med LVD, skall införas i teknisk journal för berörd flygmateriel med hänvisning till ifrågavarande LVD-nummer. Angivet underlag refererar till senaste gällande revision/utgåva. LVD utges i luftfartsverkets författningssamlingar LFS.



AIRWORTHINESS DIRECTIVE

FLIGHT STANDARDS SERVICE
REGULATORY SUPPORT DIVISION
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U.S. Department
of Transportation
**Federal Aviation
Administration**

The following Airworthiness Directive issued by the Federal Aviation Administration in accordance with the provisions of Federal Aviation Regulations, Part 39, applies to an aircraft model of which our records indicate you may be the registered owner. Airworthiness Directives affect aviation safety and are regulations which require immediate attention. You are cautioned that no person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of the Airworthiness Directive (reference FAR Subpart 39.3).

94-24-04 McDONNELL DOUGLAS HELICOPTER SYSTEMS (MDHS) and HUGHES HELICOPTERS, INC. (HUGHES): Amendment 39-9077. Docket No. 94-SW-13-AD.

Applicability: Model 369D, E, F, and FF series helicopters with pitch control assembly, part number (P/N) 369D21800 or P/N 369D21820, installed, certificated in any category.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of the inner tang of the pitch control assembly lockwasher (lockwasher), loss of the locknut, disengagement of the pitch control assembly, loss of tail rotor control, and subsequent loss of control of the helicopter, accomplish the following:

(a) Within 25 hours time-in-service or 90 calendar days after the effective date of this AD, whichever occurs first, remove the tail rotor (T/R) assembly and pitch control assembly from the T/R gearbox in accordance with the applicable maintenance manual.

(b) Inspect the lockwasher, P/N MS172209, for dents in either side of the inner tang inside radius as shown in Figure 1 of MDHS Service Information Notice (SIN) No. DN-185, EN-78, and FN-64, dated September 23, 1994, using a 5x or higher magnifying glass.

(c) Apply a 0.125 inch-wide torque stripe to the surface of the locknut and swashplate in accordance with paragraph B and C of Part II of MDHS SIN No. DN-185, EN-78, and FN-64, dated September 23, 1994, and reinstall the T/R assembly and pitch control assembly into the T/R gearbox in accordance with the applicable maintenance manual.

(d) Inspect the torque stripe for slippage at intervals not to exceed 100 hours time-in-service. If any slippage is detected, replace the lockwasher with an airworthy lockwasher in accordance with the applicable maintenance manual. Reapply the 0.125 inch-wide torque stripe to the surface of the locknut and swashplate.

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used when approved by the Manager, Los Angeles Aircraft Certification Office, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Los Angeles Aircraft Certification Office.

NOTE: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Los Angeles Aircraft Certification Office.

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(g) The inspections shall be done in accordance with McDonnell Douglas Helicopter Systems Service Information Notice No. DN-185, EN-78, and FN-64, dated September 23, 1994. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from McDonnell Douglas Helicopter Systems, Technical Publications, Bldg. 543/B111, 5000 E. McDowell Road, Mesa, Arizona 85205-9797. Copies may be inspected at the FAA, Office of the Assistant Chief Counsel, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(h) This amendment becomes effective on December 13, 1994.

FOR FURTHER INFORMATION CONTACT:

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