

Sektion 2. Utlandstillverkad flygmateriel

<u>TITEL:</u>	Kontroll / modifiering av motorfundament
<u>GÄLLER:</u>	Modellerna 95-C55, 95-C55A, D55, D55A, E55 och E55A, S/N TC 350 och S/N TE-1 t o m TE 1201. Modellerna 58 och 58A, S/N TH-1 t o m TH 1610.
<u>ÅTGÄRD:</u>	För att undvika motorvibrationer och möjlig motorseparation som följd, utför åtgärder i enlighet med bifogad kopia av FAA AD 91-15-20 och Beech Service Bulletin (SB) Nr. 2362 revision 1, daterad februari 1991 eller senare utgåva.
<u>TID FÖR ÅTGÄRD:</u>	Vid uppnådda 600 flygtimmar eller inom 100 flygtimmar räknat från detta LVD's utgivningsdatum, vilket som senast inträffar och därefter i intervaller angivna i FAA AD 91-15-20. Åtgärder enligt (3) i FAA AD 91-15-20 upphäver den återkommande inspektionen.
<u>UNDERLAG:</u>	FAA AD 91-15-20 Beech Service Bulletin (SB) Nr. 2362 revision 1, daterad februari 1991 eller senare utgåva.
<u>REFERENS:</u>	FAA AD 91-15-20
<u>UTGIVNINGS- DATUM:</u>	1991-09-19

LFS: 1991:20

Åtgärd enligt LVD utgör nödvändig förutsättning för ifrågavarande flygmateriels luftvärdighet. Referens BCL M 1.11. Anteckning om åtgärd, som vidtagits i enlighet med LVD, skall införas i teknisk journal för berörd flygmateriel med hänvisning till ifrågavarande LVD-nummer. Angivet underlag refererar till senaste gällande revision/utgåva. LVD utges i luftfartsverkets författningssamlingar LFS.



U.S. Department
of Transportation
**Federal Aviation
Administration**

AIRWORTHINESS DIRECTIVE

AVIATION STANDARDS NATIONAL FIELD OFFICE
P.O. BOX 26460
OKLAHOMA CITY, OKLAHOMA 73125

The following Airworthiness Directive issued by the Federal Aviation Administration in accordance with the provisions of Federal Aviation Regulations, Part 39, applies to an aircraft model of which our records indicate you may be the registered owner. Airworthiness Directives affect aviation safety. They are regulations which require immediate attention. You are cautioned that no person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of the Airworthiness Directive (FAR 39.3).

91-15-20 BEECH: Amendment 39-7084. Docket No. 91-CE-08-AD.

Applicability: Models 95-C55, 95-C55A, D55, D55A, E55, and E55A airplanes (serial number (S/N) TC350, and S/N TE-1 through TE-1201), and Models 58 and 58A airplanes (S/N TH-1 through TH-1610), certificated in any category.

Compliance: Required as indicated, unless already accomplished.

To prevent severe engine vibration and possible separation of the engine from the airplane caused by cracked engine mounts, accomplish the following:

(a) Upon the accumulation of 600 hours time-in-service (TIS) or within 100 hours TIS after the effective date of this AD, whichever occurs later, inspect each engine mount in accordance with the instructions in Beech Service Bulletin (SB) No. 2362, Revision 1, dated February 1991.

(1) If no cracks are found, repeat the inspection on each engine mount thereafter at intervals not to exceed 100 hours TIS.

(2) If a crack is found on an engine mount, prior to further flight, remove the engine from the affected engine mount, remove the mount from the airplane, and magnetic particle inspect the engine mount to determine the length of the crack in accordance with the instructions in Beech SB No. 2362, Revision 1, dated February 1991.

(i) If the length of the crack is .52 inches (true) or less, repair and reinforce the engine mount using Beech Kit 58-9007-1S in accordance with the instructions in Beech SB 2362, Revision 1, dated February 1991.

(ii) If the length of the crack is greater than .52 inches (true), replace the cracked engine mount with a part number (P/N) 96-910010-67 engine mount in accordance with the instructions in Beech SB No. 2362, Revision 1, dated February 1991.

(3) The repetitive inspections specified in paragraph (a)(1) of this AD may be terminated on an engine mount that has been repaired and reinforced with Beech Kit 58-9007-1S or if a P/N 96-910010-67 engine mount has been installed.

(b) Special flight permits may be issued in accordance with FAR 21.197 and 21.199 to operate airplanes to a location where the requirements of this AD can be accomplished.

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(c) An alternative method of compliance or adjustment of the initial or repetitive compliance times that provides an equivalent level of safety may be approved by the Manager, Wichita Aircraft Certification Office, FAA, 1801 Airport Road, Mid-Continent Airport, Wichita, Kansas 67209. The request should be forwarded through an appropriate FAA Maintenance Inspector, who may add comments and then send it to the Manager, Wichita Aircraft Certification Office.

(d) The inspections, modifications, and replacements required by this AD shall be done in accordance with Beech Service Bulletin No. 2362, Revision 1, dated February 1991. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR Part 51. Copies may be obtained from the Beech Aircraft Corporation, P.O. Box 85, Wichita, Kansas 67201-0085. Copies may be inspected at the FAA, Central Region, Office of the Assistant Chief Counsel, Room 1558, 601 E. 12th Street, Kansas City, Missouri, or at the Office of the Federal Register, 1100 L Street, NW, Room 8401, Washington, DC.

This amendment (39-7084, AD 91-15-20) becomes effective on September 3, 1991.

FOR FURTHER INFORMATION CONTACT:

Mr. Larry Engler, Aerospace Engineer, Wichita Aircraft Certification Office, FAA, 1801 Airport Road, Mid-Continent Airport, Wichita, Kansas 67209; Telephone (316) 946-4409.