
Sektion 2. Utlandstillverkad flygmateriel

TITEL: Kontroll/byte av höjdrimdomkraft

GÄLLER: Alla SA226 och SA227 (alla modeller och S/N) med installerad höjdrimdomkraft Simmonds-Precision P/N DL5040M5, P/N DL5040M6 eller P/N DL5040M8, Barber-Colman P/N 27-19008-00-001, P/N 27-19008-002, P/N 27-19008-00-004 eller P/N 27-19008-005.

ÅTGÄRD: Utför åtgärder angivna i bifogad kopia av FAA AD 2000-03-17

TID FÖR ÅTGÄRD: I enlighet med FAA AD 2000-03-17 räknat från detta LVD's utgivningsdatum

UNDERLAG: FAA AD 2000-03-17 och däri angivet underlag

REFERENS: FAA AD 2000-03-17

BESLUTSDATUM: 2000-03-08

LFS 2000:32

Åtgärder enligt LVD utgör nödvändig förutsättning för ifrågavarande flygmateriels luftvärdighet. Referens BCL M 1.11.
Anteckning om åtgärd, som vidtagits i enlighet med LVD, skall införas i teknisk journal för berörd flygmateriel med hänvisning till ifrågavarande LVD-nummer. Angivet underlag refererar till senast gällande revision/utgåva. LVD utges i luftfartsverkets författningssamlingar LFS.

Postadress	Gatuadress	Telefonnummer	Faxnummer	Telegram Civilair	Telex
601 79 NORRKÖPING	Vikobplan 11	011-19 20 00	011-19 25 75	Norrköping	62450



AIRWORTHINESS DIRECTIVE

REGULATORY SUPPORT DIVISION
P.O. BOX 26460
OKLAHOMA CITY, OKLAHOMA 73125-0460

U.S. Department
of Transportation
**Federal Aviation
Administration**

The following Airworthiness Directive issued by the Federal Aviation Administration in accordance with the provisions of Title 14 of the Code of Federal Regulations (14 CFR) part 39, applies to an aircraft model of which our records indicate you may be the registered owner. Airworthiness Directives affect aviation safety and are regulations which require immediate attention. You are cautioned that no person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of the Airworthiness Directive (reference 14 CFR part 39, subpart 39.3).

2000-03-17 FAIRCHILD AIRCRAFT, INC.: Amendment 39-11576; Docket No. 99-CE-59-AD, Supersedes AD 97-23-01, Amendment 39-10188; which superseded AD 93-15-02 R2, Amendment 39-9689; which revised AD 93-15-02 R1, Amendment 39-9180; which revised AD 93-15-02, Amendment 39-8648.

Applicability: All SA226 and SA227 series airplanes (all models and serial numbers), certificated in any category.

NOTE 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated in the body of this AD, unless already accomplished.

To detect excessive freeplay or rod slippage in the pitch trim actuator, which, if not detected and corrected, could result in pitch trim actuator failure and possible loss of control of the airplane, accomplish the following:

NOTE 2: The paragraph structure of this AD is as follows:

Level 1: (a), (b), (c), etc.

Level 2: (1), (2), (3), etc.

Level 3: (i), (ii), (iii), etc.

Level 2 and Level 3 structures are designations of the Level 1 paragraph they immediately follow.

(a) Accomplish the following at the times specified in the chart in paragraph (b) of this AD:

(1) Initial and repetitive inspections:

(i) For airplanes equipped with a Simmonds-Precision actuator, P/N DL5040M5, P/N DL5040M6, or P/N DL5040M8, measure the freeplay (inspection) of the pitch trim actuator and inspect the actuator for rod slippage in accordance with the INSTRUCTIONS section of Fairchild Aircraft SA226 Series Service Letter (SL) 226-SL-005, or Fairchild Aircraft SA227 Series SL 227-SL-011, both Revised: August 3, 1999; or Fairchild Aircraft SA227 Series Service Letter CC7-SL-028, Issued: August 12, 1999, as applicable.

(ii) For airplanes equipped with Barber-Colman actuators, P/N 27-19008-00-001, P/N 27-19008-002, P/N 27-19008-00-004, or P/N 27-19008-005, conduct a functional inspection of the actuator in accordance with the INSTRUCTIONS section of Fairchild Aircraft SA226 Series SL 226-SL-014, Revised: February 1, 1999, Fairchild Aircraft SA227 Series SL 227-SL-031, Revised: February 1, 1999, or Fairchild Aircraft SA227 Series SL CC7-SL-021, Revised: February 1, 1999, whichever is applicable.

NOTE 3: The actions in this AD are the same as the actions in AD 97-23-01, except for the actions added to the airplanes equipped with improved design pitch trim actuators.

(2) Initial and repetitive replacements: Replace the pitch trim actuator with any of the pitch trim actuators presented in the Chart in paragraph (b) of this AD, as applicable, at the time specified in the Repetitive Replacement column of this chart. However, if certain freeplay limitations that are specified in the service letters are exceeded or if rod slippage is found, prior to further flight, replace the pitch trim actuator.

(b) The following chart presents the pitch trim actuator that could be installed and the initial and repetitive inspection and replacement compliance times of this AD:

Condition	Initial Inspection	Repetitive Inspection	Repetitive Replacement
For all affected airplane models, except for the Models SA227-CC and SA227-DC, with an original Simmonds-Precision actuator, P/N DL5040M5, installed.	Upon accumulating 3,000 hours TIS on a Simmonds-Precision P/N DL5040M5 actuator or within 50 hours TIS after April 17, 1995 (the effective date of AD 93-15-02 R1), whichever occurs later.	Every 250 hours TIS after the initial inspection until accumulating 5,000 hours TIS on the actuator or 500 hours TIS after the last inspection required by AD 93-15-02 R1, whichever occurs later.	Initially upon accumulating 5,000 hours TIS on the actuator or 500 hours TIS after the initial inspection, whichever occurs later, and thereafter as indicated below.

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Condition	Initial Inspection	Repetitive Inspection	Repetitive Replacement
For all affected airplane models, except for the Models SA227-CC and SA227-DC, with a replacement Simmonds-Precision actuator, P/N DL5040M5, installed.	Initially upon accumulating 5,000 hours TIS on the new actuator or within 50 hours TIS after April 17, 1995 (the effective date of AD 93-15-02 R1), whichever occurs later.	Every 300 hours TIS after the initial inspection until accumulating 6,500 hours TIS on the actuator.	Upon accumulating 6,500 hours TIS on the actuator.
For all affected airplane models, except for the Models SA227-CC and SA227-DC, with a replacement Simmonds-Precision actuator, P/N DL5040M6, installed. This part can be new, modified from a P/N DL5040M5 actuator, or overhauled and zero-timed.	Initially upon accumulating 7,500 hours TIS on the new or modified actuator or within 50 hours TIS after April 17, 1995 (the effective date of AD 93-15-02 R1), whichever occurs later.	Every 300 hours TIS after the initial inspection until accumulating 9,900 Hours TIS on the actuator.	Upon accumulating 9,900 hours TIS on the actuator.
For all affected airplane models, except for the Models SA227-CC and SA227-DC, with a replacement Simmonds-Precision actuator, P/N DL5040M5, installed that was overhauled and zero-timed where both nut assemblies, P/N AA56142, were replaced with new assemblies during overhaul.	Initially upon accumulating 5,000 hours TIS on the over-hauled actuator or within 50 hours TIS after April 17, 1995 (the effective date of AD 93-15-02 R1), whichever occurs later.	Every 300 hours TIS after the initial inspection until Accumulating 6,500 hours TIS on the actuator.	Upon accumulating 6,500 hours TIS on the actuator.
For all affected airplane models, except for the Models SA227-CC and SA227-DC, with a replacement P/N DL5040M5 actuator installed that was overhauled and zero-timed where both nut assemblies, P/N AA56142, were not replaced with new assemblies during overhaul.	Initially upon accumulating 3,000 hours TIS on the over-hauled actuator or within 50 hours TIS after April 17, 1995 (the effective date of AD 93-15-02 R1), whichever occurs later.	Every 250 hours TIS after the initial inspection until accumulating 5,000 hours TIS on the actuator.	Upon accumulating 5,000 hours TIS on the actuator.
For all affected airplane models with a newly fabricated or overhauled and zero-timed Barber-Colman actuator, P/N 27-19008-001/-004 or P/N 27-19008-002/-005.	Upon accumulating 500 hours total TIS on the newly fabricated or over-hauled and zero-timed actuator or within 50 hours TIS after the effective date of AD 97-23-01, whichever occurs later.	Every 300 hours TIS after the initial inspection.	None.
For the Models SA227-CC and SA227-DC only, with a Simmonds-Precision pitch trim actuator, P/N DL5040M5 or P/N DL5040M6, installed	None.	None.	Upon accumulating 1,500 hours TIS on the actuator.
For all affected airplanes with a Barber-Colman P/N 27-19008-006 or 27-19008-007 actuator installed.	Must be overhauled upon the accumulation of 2,000 hours TIS on the actuator	Must be overhauled at intervals not to exceed 2,000 hours TIS.	No replacement requirements.
For all affected airplanes with a Simmonds-Precision pitch trim actuator, P/N DL5040M8, installed	Upon accumulating 7,500 hours TIS on the actuator or within the next 50 hours TIS after the effective date of this AD, whichever occurs later.	Every 600 hours TIS after the initial inspection until accumulating 9,900 hours TIS.	Upon accumulating 9,900 hours TIS on the actuator.

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(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(d) An alternative method of compliance or adjustment of the initial or repetitive compliance times that provides an equivalent level of safety may be approved by the Manager, Airplane Certification Office (ACO), FAA, 2601 Meacham Boulevard, Fort Worth, Texas 76193-0150.

(1) The request shall be forwarded through an appropriate FAA Maintenance Inspector, who may add comments and then send it to the Manager, Fort Worth Airplane Certification Office.

(2) Alternative methods of compliance that were approved in accordance with AD 97-23-01 are considered to be approved as alternative methods of compliance with this AD.

NOTE 4: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Fort Worth Airplane Certification Office.

(e) (1) The inspections required by this AD shall be done in accordance with the following:

(i) Fairchild Aircraft SA226 Series SL 226-SL-005, Revised: August 3, 1999; or

(ii) Fairchild Aircraft SA227 Series SL 227-SL-011; Revised: August 3, 1999; or

(iii) Fairchild Aircraft SA227 Series SL CC7-SL-028, Issued: August 12, 1999; and

(iv) Fairchild Aircraft SA226 Series SL 226-SL-014, Revised: February 1, 1999; or

(v) Fairchild Aircraft SA227 Series SL 227-SL-031, Revised: February 1, 1999; or

(vi) Fairchild Aircraft SA227 Series SL CC7-SL-021, Revised: February 1, 1999.

(2) This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Field Support Engineering, Fairchild Aircraft Inc., P.O. Box 790490, San Antonio, Texas 78279-0490. Copies may be inspected at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 301, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

(f) This amendment supersedes 97-23-01, Amendment 39-10188; which superseded AD 93-15-02 R2, Amendment 39-9689; which revised AD 93-15-02 R1, Amendment 39-9180; which revised AD 93-15-02, Amendment 39-8648.

(g) This amendment becomes effective on April 10, 2000.

FOR FURTHER INFORMATION CONTACT:

Mr. Werner Koch, Aerospace Engineer, FAA, Airplane Certification Office, 2601 Meacham Boulevard, Fort Worth, Texas 76193-0150; telephone: (817) 222-5133; facsimile: (817) 222-5960.

Issued in Kansas City, Missouri, on February 9, 2000.

Michael K. Dahl, Acting Manager, Small Airplane Directorate, Aircraft Certification Service