
Sektion 2. Utlandstillverkad flygmateriel

TITEL: Sprickkontroll av lastdörrens spant

GÄLLER: Modellerna SA226-AT, SA226-TC, SA227-AC och
SA 227-AT med S/N angivna i bifogad kopia av FAA AD 98-06-25

ÅTGÄRD: Utför åtgärder angivna i FAA AD 98-06-25

TID FÖR ÅTGÄRD: Inom 500 flygtimmar räknat från 27 april 1998

UNDERLAG: FAA AD 98-06-25 och där angivet tillämpligt underlag

REFERENS: FAA AD 98-06-25

BESLUTSDATUM: 1998-04-24

LFS 1998:28

Åtgärder enligt LVD utgör nödvändig förutsättning för ifrågavarande flygmateriels luftvärdighet. Referens BCL M 1.11.

Anteckning om åtgärd, som vidtagits i enlighet med LVD, skall införas i teknisk journal för berörd flygmateriel med hänvisning till ifrågavarande LVD-nummer. Angivet underlag refererar till senast gällande revision/utgåva. LVD utges i luftfartsverkets författningssamlingar LFS.

Postadress	Gatuadress	Telefonnummer	Telegram	Telex
601 79 NORRKÖPING	Vikboplan 11	011-192000	Civilair Norrköping	62450

AIRWORTHINESS DIRECTIVE

Bilaga till LVD 2861

REGULATORY SUPPORT DIVISION
P.O. BOX 26460
OKLAHOMA CITY, OKLAHOMA 73125-0460

U.S. Department
of Transportation
Federal Aviation
Administration

The following Airworthiness Directive issued by the Federal Aviation Administration in accordance with the provisions of Federal Aviation Regulations, Part 39, applies to an aircraft model of which our records indicate you may be the registered owner. Airworthiness Directives affect aviation safety and are regulations which require immediate attention. You are cautioned that no person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of the Airworthiness Directive (reference FAR Subpart 39.3).

98-06-25 FAIRCHILD AIRCRAFT INC.: Amendment 39-10403; Docket No. 96-CE-68-AD.

Applicability: The following models and serial numbered airplanes, certificated in any category.

Models	Serial Numbers
SA226-AT	AT001 through AT074
SA226-TC	TC201 through TC419
SA227-AC	AC406, AC415, AC416, AC420 through AC456, AC458 through AC469, and AC471 through AC478
SA227-AT	AT423 through AT469

NOTE 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated within the body of this AD, unless already accomplished.

To prevent failure of the cargo door in flight, which, if not corrected, could cause decompression injuries to passengers and substantial structural damage to the airplane, accomplish the following:

(a) Within the next 500 hours time-in-service (TIS) after the effective date of this AD, inspect the cargo door lower belt frames at the cargo latch receptacles for cracks in accordance with Part A of the ACCOMPLISHMENT INSTRUCTIONS section in Fairchild Aircraft SA226 Series Service Bulletin (SB) No. 226-53-007, Issued: May 7, 1981; Revised: February 17, 1992, or Fairchild Aircraft SA227 Series SB No. 227-53-003, Issued: January 29, 1986; Revised: February 13, 1986, whichever is applicable.

(b) If cracks are found during the inspection required in paragraph (a) of this AD, prior to further flight, accomplish the following:

(1) For belt frames located at Fuselage Station (F.S.) 438.060 and F.S. 491.060, repair the belt frame by installing angle part number (P/N) 27-22206-009 or P/N 27-22206-010, in accordance with the Fairchild Aircraft SA226/227 Structural Repair Manual (SRM), Section 53-90-20, Initial Issue: March 1, 1983, Revision 24, dated August 27, 1997; or, Fairchild Aircraft Approved Repair Procedure (ARP) 53-30-9701, dated July 28, 1997. The reinforcement doublers (P/N 27-22206-007 and -008) are also needed together with this repair.

(2) For belt frames located at F.S. 454.501, F.S. 455.726, F.S. 473.392, and F.S. 474.657, replace all four belt frames with new design frames, P/N 27-22207-008, 27-22208-005, 27-22208-005, and 27-22207-007, respectively, in accordance with the Fairchild Aircraft SA226/227 SRM, Section 53-90-20, Initial Issue: March 1, 1983, Revision 24, dated August 27, 1997; or, Fairchild Aircraft ARP 53-30-9701, dated July 28, 1997. No reinforcement doublers are needed for these four new design belt frames.

(c) If no cracks are found in all six belt frames during the inspection required by paragraph (a) of this AD, install reinforcement doublers in all six belt frames within 500 hours TIS from the initial inspection, in accordance with Part B of the ACCOMPLISHMENT INSTRUCTIONS of Fairchild Aircraft SA226 Series Service Bulletin (SB) No. 226-53-007, Issued: May 7, 1981; Revised: February 17, 1992, or Fairchild Aircraft SA227 Series SB No. 227-53-003, Issued: January 29, 1986; Revised: February 13, 1986, whichever is applicable.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(e) An alternative method of compliance or adjustment of the initial or repetitive compliance time that provides an equivalent level of safety may be approved by the Manager, Fort Worth Airplane Certification Office, 2601 Meacham Boulevard, Fort Worth, Texas 76193-0150. The request shall be forwarded through an appropriate FAA Maintenance Inspector, who may add comments and then send it to the Manager, Fort Worth Airplane Certification Office.

NOTE 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Manager, Fort Worth Airplane Certification Office.

(f) The inspections and modifications required by this AD shall be done in accordance with the following service information:

- Fairchild Aircraft Corporation SA227 Series Service Bulletin No. 227-53-003, Issued: January 29, 1986, Revised: February 13, 1986,
- Fairchild Aircraft Corporation SA226 Series Service Bulletin No. 226-53-007, Issued: May 7, 1981, Revised: February 17, 1992,
- Fairchild Aircraft SA226/SA227 Structural Repair Manual (SRM) section 53-90-20, Initial Issue: March 1, 1983, Revision 24, dated August 27, 1997, and
- Fairchild Aircraft Approved Repair Procedure (ARP) 53-30-9701, dated July 28, 1997.

This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Fairchild Aircraft, P.O. Box 790490, San Antonio, Texas 78279-0490, telephone (210) 824-9421. Copies may be inspected at the FAA, Central Region, Office of the Regional Counsel, Room 1558, 601 E. 12th Street, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

(g) This amendment (39-10403) becomes effective on April 27, 1998.

FOR FURTHER INFORMATION CONTACT:

Mr. Hung Viet Nguyen, Aerospace Engineer, FAA, Fort Worth Airplane Certification Office, 2601 Meacham Boulevard, Fort Worth, Texas 76193-0150; telephone (817) 222-5155; facsimile (817) 222-5960.